

VELOCITY INCREMENT REQUIRED TO CIRCULARIZE AN ORBIT

Assume that a spacecraft is in a non-circular orbit with semi-major axis, a , and eccentricity, e . Suppose that it is required that the orbit be circularized at a radius, r , such $r_p < r < r_a$ using a single burn (we will consider the planar case only). The magnitude of velocity in the non-circular orbit at the radius where the orbit is to be circularized is given by

$$V = \sqrt{\mu \left(\frac{2}{r} - \frac{1}{a} \right)}$$

Recall that the orbit parameter (semi-latus rectum) of the orbit is given by the relation

$$p = a(1 - e^2). \quad \text{The true anomaly, } f, \text{ is then given by } f = \cos^{-1} \frac{p - r}{e r}.$$

(For this calculation, do not worry about obtaining the proper quadrant for f , use the value between zero and 180 degrees which is output by your calculator. The velocity increment required is the same for both possible values of the angle, f . If you need to know the proper value of f , recall that on the outbound half of the orbit, f lies between zero and 180 degrees whereas on the inbound half, f lies between 180 degrees and 360 degrees.

At any point in the orbit, velocity is given by the vector equation

$$\bar{V} = \sqrt{\frac{\mu}{p}} e \sin f \bar{u}_r + \frac{\sqrt{\mu p}}{r} \bar{u}_h$$

where \bar{u}_h is a unit vector in the horizontal direction, \bar{u}_r is a unit vector in the radial (outward) direction, and μ is the gravitational parameter of the central body.

At any radius, the required circular velocity is given by $\bar{V}_C = \sqrt{\frac{\mu}{r}} \bar{u}_h$

The \bar{V} required for circularization is given by $\bar{V} = \bar{V}_C - \bar{V}$, or in vector terminology, by

$$\bar{V} = \sqrt{\frac{\mu}{r}} \bar{u}_h - \frac{\sqrt{\mu p}}{r} \bar{u}_h - \sqrt{\frac{\mu}{p}} e \sin f \bar{u}_r$$

The magnitude of the required one-burn \bar{V} is then given by

$$|\bar{V}| = \sqrt{\left(\sqrt{\frac{\mu}{r}} - \frac{\sqrt{\mu p}}{r} \right)^2 + \frac{\mu}{p} e^2 \sin^2 f}$$